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GENERAL MOTORS CORP INDIANAPOLIS IN DETROIT DIESEL A--ETC F/6 21/5
ALLISON PD370-41 DERIVATIVE TURBOPROP ENGINE.(U)

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of fuel consumption, weight, cost and complexity. The three turboprop engines selected for further study were as follows: a derivative of the unity size T701-AD-700 shaft power engine with rematched turbine (PD 370-37), an advanced T701 turboprop derivative with 25:1 overall pressure ratio and a scaled ATEGG demonstrated compressor (PD 370-40), an advanced T701 turboprop derivative with 17.7:1 overall pressure ratio and a scaled ATEGG demonstrated compressor (PD 370-41). Data is also presented on a new advanced turboprop engine with 30:1 overall pressure ratio which incorporates compressor, combustor, turbine, and cooling technology now under development and demonstration at DDA. The documentation consists of six separate reports prepared in the following manner. One report summarizes the engine screening analysis and describes the approach to, and the conclusions of the study. A separate report for each of the three derivative engines and for the new turboprop present estimates of performance, weight, and dimensional data. The engineering budgetary estimates of the development, acquisition, and service costs for each of the four engines are presented in a separate report.

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REVISIONS

<u>Letter</u>	<u>Page</u>	<u>Revision</u>
A	4	Gearbox and total weight
A	6	Gearbox, interconnecting struts and shaft, and total weight
A	10	Additional matrix points at 0 and 25,000 feet
A	13	Additional performance
A	18	Additional performance
A	23	Additional performance
A	28	Additional performance



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I. INTRODUCTION

This report presents estimates of performance, weight, and dimensional data for the PD 370-41 turboprop engine. The PD 370-41 represents an advanced T701 turboprop derivative engine with 17.7:1 overall pressure ratio. It incorporates a scaled ATEGG demonstrated compressor with basic shaft, bearing and turbine arrangements from the new T701 turboshaft engine. The engine is in the 10,000 to 11,000 SHP class. The data is submitted for use in preliminary design type studies in the evaluation of turboprop systems.

The basic T701 engine is a free turbine turboshaft engine that was developed through safety demonstration testing, for the U.S. Army's HLH program. The model 570, a commercial industrial version of the T701, has undergone additional development testing, and is now in production.

The reduction gearbox for speed reduction to the prop-fan is a new simplified design, compared to the DDA T56 series of gearboxes. The new design is based upon a study into the reliability and maintenance cost history of past turboprop systems, and follows the recommendations of that study for a gearbox with high reliability, easy maintainability, and low maintenance costs.



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II. ENGINE DESCRIPTION

The Model PD 370-41 is an axial flow engine, having a single spool core and a free power turbine connected by shafting, and supporting structure to an offset reduction gear assembly. The general arrangement and external features of the engine are shown in Figure II-1, with principle physical characteristics listed in Table II-I. Output speed of the engine is constant at 10,600 RPM. The reduction gearbox shown in Figure II-1 has an overall gear ratio of 8.52:1, providing a propfan speed of 1244 rpm at 10,600 engine rpm. Parametric weight data is shown in Section III so that other propfan rotational speeds, and gear ratios can be analyzed. An aircraft accessory drive pad is provided on the back of the gearbox to drive an aircraft mounted accessory drive box. Power available at this pad is 500 HP at 8000 rpm. The primary engine mounts are on the gearbox with a hang mount at the rear of the engine. Engine accessories are driven by a bevel drive from the high pressure spool. The control system is integral with the prop-fan and is digital electronic. The oil system is integral to the engine and also supplies the prop-fan and reduction gearbox, but is separately filtered and monitored to isolate fault detection in each of these major modules. Engine torque is measured hydraulically from the gear thrust of the power train idler gears in the reduction gearbox.

The gearbox is shown offset, based upon DDA's experience with large turbo-prop engines. It is offset up to be consistent with current studies showing a preference to under-the-wing engine mounting. It can also be supplied in the offset-down position.

Performance ratings, sea level static, are listed in Table II-II.



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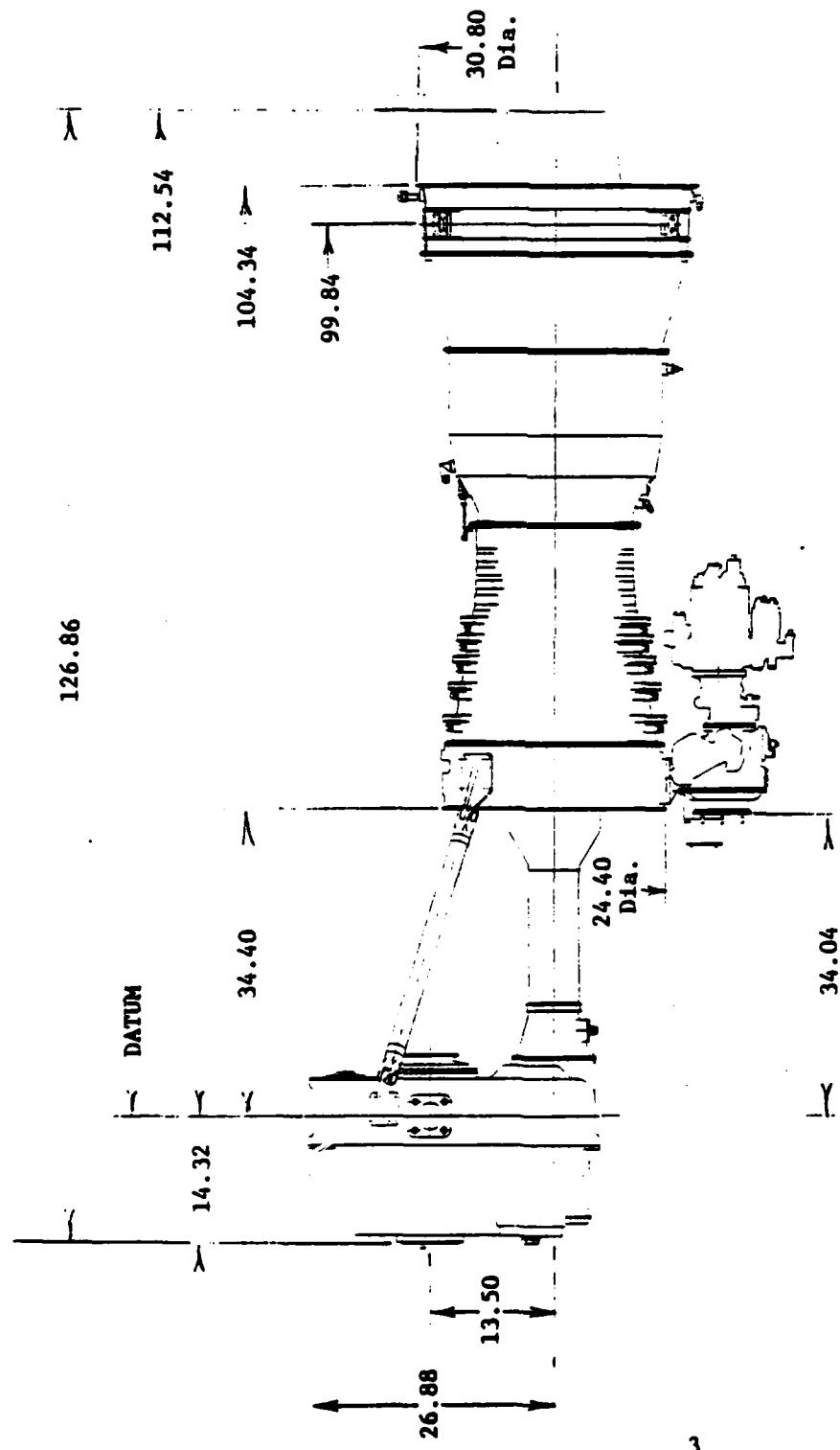


Figure II-1. PD370-41 General Arrangement



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TABLE II-I

PD 370-41 PHYSICAL CHARACTERISTICS

(Includes Gearbox)

Length (in)	126.86
Max. Engine Diameter (in)	30.80
Max. Gearbox Offset, upward (in)	26.88
Dry Weight, lbs	
Engine	1503
Gearbox, including interconnecting struts and shaft	634
Total	2137

For gear ratios other than 8.52:1 the reduction gearbox dimensions may be scaled as follows:

$$\text{Dimensions} = \text{Base dim.} \times \left(\frac{\text{GR}}{8.52} \right)^{0.33}$$



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TABLE II-II

PD370-41 PERFORMANCE SUMMARY

Sea Level, 0 kts

	<u>Standard Day</u>			<u>Hot Day, 89.8° F</u>		
	SHP	SFC	F _N	SHP	SFC	F _N
Take-Off	10,974	0.389	1,247	9,114	0.407	997
Max. Continuous	8,133	0.409	908	6,713	0.432	738



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III. WEIGHTS

The weight of the engine, gearbox, and the interconnecting struts and shaft are given in Table III-I. The gearbox weight is based upon a gear ratio of 8.52:1 which provides a propfan speed of 1244 rpm.

TABLE III-I
PD 370-41 WEIGHTS

	<u>Dry</u>	<u>Wet*</u>	<u>Installed</u>
Basic Engine, lbs	1503	1526	1526
Gearbox, lbs	598	641	641
Interconnecting Struts and Shaft, lbs	36	36	36
Total, lbs	2137	2203	2203

* Includes total amount of oil required for engine and gearbox operation.

For gear ratios other than 8.52:1, the gearbox, interconnecting strut, and shaft dry weight may be estimated as follows:

$$\text{Dry gearbox weight} = 598 \times \left(\frac{\text{GR}}{8.52} \right)^{0.4}$$

Interconnecting struts and shaft weight = 6.0% of dry gearbox weight



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IV. STEADY STATE PERFORMANCE

Steady state performance data is tabulated in this section for all points shown in Figure IV-1. Basic engine data is shown for the following assumptions:

- o Uninstalled engine
- o ICAO standard atmosphere except for takeoff which in addition includes an ambient temperature of 89.8° F at standard atmosphere
- o 100% inlet recovery
- o Zero accessory horsepower extraction
- o Zero customer bleed extraction
- o Zero losses due to reduction gear
- o Fuel heating value - 18,400 Btu/lb
- o Estimated average engine performance - No SHP or fuel flow guarantee factors

Sensitivity data is provided for each point so that bleed and duct losses may be estimated as required.

Nomenclature

Nomenclature used in the tabulation of performance is as follows:

MACH	Mach number
SHP	Shaft horsepower
SFC	Specific fuel consumption, lbs/hr/hp
WF	Engine fuel flow, lbs/hr
FN	Net jet thrust, lbs (jet gross thrust - ram drag)
ESHP	Equivalent shaft horsepower (energy in jet stream converted ideally to horsepower and added to SHP)
WCIN	Total inlet corrected airflow, $W\sqrt{\theta_1}/\delta_1$

where: $\theta_1 = \frac{\text{Engine inlet total temp, } {}^{\circ}\text{R}}{518.688}$



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$$\delta_1 = \frac{\text{Engine inlet total pressure, psi}}{14.696}$$

TNOZ	Jet nozzle total temperature, °R
PNOZ	Jet nozzle total pressure, psi
RC	Compressor pressure ratio
BOT	Burner outlet temperature, °R
NO	Point number

Sensitivity Data

Bleed:

SHP, with bleed = SHP, no bleed - (DEL SHP)(% bleed)

WF, with bleed = WF, no bleed - (DEL WF)(% bleed)

FN, with bleed = FN, no bleed - (DEL FN)(% bleed)

Inlet Recovery:

η_R = Total pressure actual/Total pressure ideal

SHP, with recovery = SHP, ideal recovery - (DEL SHP)(1 - η_R)(100)

WF, with recovery = WF, ideal recovery (η_R)

FN, with recovery = FN, ideal recovery - (DEL FN)(1 - η_R)(100)

Jet Nozzle Duct Loss:

To estimate thrust loss due to additional duct loss prior to the jet nozzle, use the following equation:

FN, with loss = FN, without loss - FN, without loss (K) $\left(\frac{\Delta P}{P}\right)$

where,

- o K is obtained for each point from sensitivity data
- o $\frac{\Delta P}{P} = \frac{PTOT, \text{ no loss} - PTOT, \text{ total loss}}{PTOT, \text{ no loss}}$



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Reduction Gear Loss:

Reduction gear is 99 percent efficient.

Accessory Drive Losses:

Accessory drive power extraction is directly from the accessory drive pad on the reduction gearbox. Reduce SHP to prop-fan by amount of accessory power extraction at each point.

Nozzle Throat Area

The effective nozzle throat area is constant for all conditions at 398.4 in².



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Standard and Hot Day; Takeoff and Maximum Continuous

Altitude (Ft $\times 10^{-3}$)	0	.1	.2	.3
0	X	X	X	X

Standard Day; Maximum Climb, Maximum Continuous and Part Power to Idle

Altitude (Ft $\times 10^{-3}$)	.2	.3	.4	.5	.6	.7	.75	.8
0	X	X	X	X	X	X	X	X
5	X	X	X	X	X	X	X	X
10	X	X	X	X	X	X	X	X
15		X	X	X	X	X	X	X
20		X	X	X	X	X	X	X
25		X	X	X	X	X	X	X
30		X	X	X	X	X	X	X
35		X	X	X	X	X	X	X
40		X	X	X	X	X	X	X
45		X	X	X	X	X	X	X

Figure IV-1. Matrix of flight conditions for performance data

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PN370-41 TURBOPROP
00 PERCENT RECOVERY

GENERAL MOTORS CORPORATION **ZERO POWER EXTRACTION**

DETROIT DIESEL ALLISON DIVISION

EDR 9776

PD370-41 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

0 FEET ALTITUDE

POWER	MACH	BEEF-GEL-IMUL-BEEF	NET SHP BEEF-GEL-BEEF-NO-FR	K	NO
70. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	36.3 35.0 34.6 34.6 34.6 34.6
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	5.3 5.3 5.3 5.3 5.3 5.3
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	69.6 69.6 69.6 69.6 69.6 69.6
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	7.87 7.87 7.87 7.87 7.87 7.87
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	12.75

T AMBIENT = 89.8°F

0 FEET ALTITUDE

10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	9.0 9.0 9.0 9.0 9.0 9.0
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	10.2
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	10.8
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	11.0
10. H.C.	0.0	170.0 170.0 170.0 170.0 170.0 170.0	23.1 23.1 23.1 23.1 23.1 23.1	62.3 62.3 62.3 62.3 62.3 62.3	11.2

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PD370-41 TURBOPROP
100 PERCENT RECOVERY

GENERAL MOTORS CORPORATION

ZERO POWER EXTRACTION

100 PERCENT RECOVERY

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DETROIT DIESEL ALLISON DIVISION

GENERAL MOTORS CORPORATION

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PD370-41 TURBOPROP
100 PERCENT RECOVERY
STD DAY

ZERO POWER EXTRACTION

POWER	MACH	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	NO	NO
PN02	TNO2	PN02	TNO2	PN02	TNO2	AC	AC
5000 FEET ALTITUDE	MCIN						
FN	ESHP	FN	ESHP	FN	ESHP	FN	ESHP
WF	SFC	WF	SFC	WF	SFC	WF	SFC
SIP	MACH	SIP	MACH	SIP	MACH	SIP	MACH
CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.	CLIMB H.C.

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PD370-41 TURBOPROP
100 PERCENT RECOVERY

GENERAL MOTORS CORPORATION

ZERO STRESS

100 PERCENT RECOVERY

ZERO POWER EXTRACTION

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PN370-41 TURBOPRO
100 PERCENT RECOV
STD DAY

GENERAL MOTORS CORPORATION

ZERO POWER EXTRACTION

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100 PERCENT RECOVERY
SID DAY

GENERAL MOTORS CORPORATION
ZERO POWER EXTRACTION

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PC 370-41 TURBOPROP

100 PERCENT RECOVERY

GENERAL MINES COMMISSION

ZERO POWER EXTRACTION

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EDR 9776
PD370-41 TURBOPROP
100 PERCENT RECOVERY
STD DAY

GENERAL MOTORS CORPORATION
ZERO POWER EXTRACTION

POWER	MACH	SHP	SFC	WF	FN	ESHP	WCIN	TNOZ	PNOZ	RC	NO
CLIMB H.C.	0.50	37520	145	22967	9937	37520	5997	25298	5997	159	1
ZERO BLEED	CLIMB H.C.	37520	145	22967	9937	37520	5997	25298	5997	159	1
ZERO BLEED	CLIMB H.C.	37520	145	22967	9937	37520	5997	25298	5997	159	1
ZERO BLEED	CLIMB H.C.	37520	145	22967	9937	37520	5997	25298	5997	159	1

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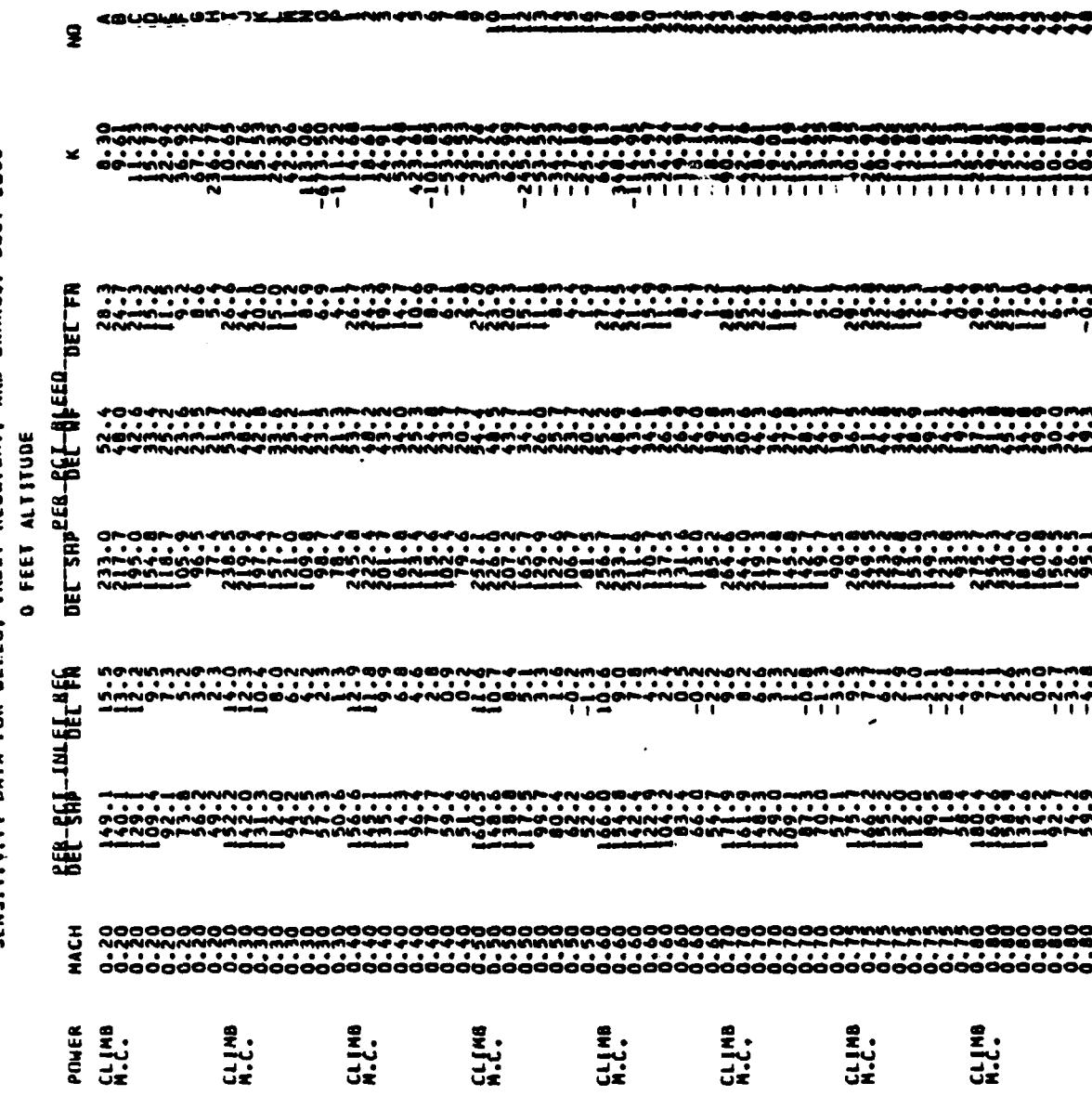
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PO370-41 TURBOPROP
100 PERCENT RECOVERY
SIX DAY

GENERAL MOTORS CORPORATION
ZERO POWER EXTRACTION

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PO370-41 TURBOPROP
SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

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PD 370-41 TURBOPROP

GENERAL MOTORS CORPORATION

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

PEP-EGF-1MLT-FBEG
MACM
POWERED
DET-STEP PEA-FGK-AP-NEED-BECKER

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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST MUFF LOSS

15000 FEET ALTITUDE

POWER CLIMB H.C.

MACH CLIMB H.C.

BLEED-INLET REC DEFL-SPIR-REC-BLEED-DEC-FR



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P0370-41 TURBOPROP
SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

20000 FEET

ALTITUDE

NET-SAP-BE[~~B~~]-FEED-DEC-FR

BEF-SAP-~~BE[~~B~~]~~-FEED-DEC-FR

BEF-SAP-~~BE[~~B~~]~~-INLET-BEF

MACH

POWER

CLIMB

H.C.

CLIMB

H.C.

CLIMB

H.C.

NO

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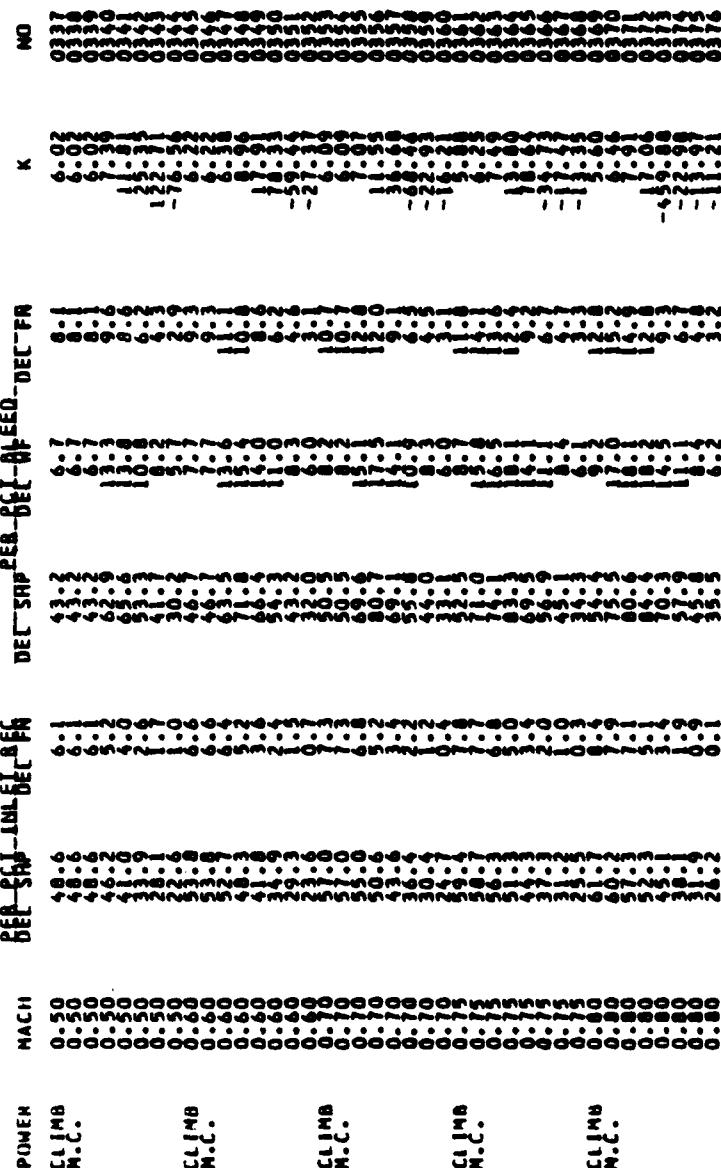
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PD370-41 TURBOPROP

SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

35000 FEET ALTITUDE

POWER - CLIMB - INLET RECOVERY - Duct Loss - DEC-TR



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SENSITIVITY DATA FOR BLEED, INLET RECOVERY, AND EXHAUST DUCT LOSS

40000 FEET ALTITUDE

POWFR	CLIMA G.I.C.	CLIMA H.I.G.	CLIMA N.I.G.	CLIMA R.I.G.
MACH	0.6000000000000000	0.6000000000000000	0.6000000000000000	0.6000000000000000
BLEED-FAIR-IMBAL-BLEED	0.6000000000000000	0.6000000000000000	0.6000000000000000	0.6000000000000000
BLEED-SAF-PER-BLEED-IMBAL-BLEED-FR	0.6000000000000000	0.6000000000000000	0.6000000000000000	0.6000000000000000
NO	0.6000000000000000	0.6000000000000000	0.6000000000000000	0.6000000000000000
K	0.6000000000000000	0.6000000000000000	0.6000000000000000	0.6000000000000000

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CONFIDENTIAL DATA FOR RACED AMMET RECOVERY AND EXAMINEE DUTY 1-155

EED INI ET RECOVERD

CLIMB RATE

POWER

NO

K

ALTITUDE

GEAR-DOWN

GEAR-UP

DEET-FR

0.5

0.0

100

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0

